

# The Impact Behaviour of Carbon Fiber-Epoxy Composite Leading Edge using Finite Element Method

O.Nurihan, Y.Aminanda, J.S.M Ali, Z.A. Rasid, Y. Hafizal

**Abstract:** Composite material has been widely used in aircrafts due to its high strength to weight ratio that leads to weight saving of the aircrafts. Equally important, aircraft material should be tough i.e. it should have the ability to absorb high energy and thus resist fracture. The aircraft's wing design requires the material to have high toughness as parts of the wing especially its leading edge is subjected to impact loadings. Using finite element software of LS-DYNA, this research focuses on studying the impact behaviour of composite panels that represent the leading edges of wings when the panels are subjected to rigid sphere projectile. Three shapes of panels are used: flat, semi-circular and semi ellipse while panels can be of 2, 4 and 8 layers to vary its thickness. The panels are made of laminated composites with woven carbon fibres and the angle of orientations are  $[0/90]_m$ ,  $[0/45]_n$  and  $[45/-45]_n$  where  $n$  will give the number of layer for the composite. The Mat-58 material type suitable for woven type fibre is used where failure criteria of Hashin is applied. It was found that the simulation results are in a very close agreement with the finding from experiments conducted earlier. Furthermore, the optimum stacking sequence was found to be the  $[0/45]_2$  stacking sequences

**Index Terms:** Energy absorption, FEA Simulation, Impact loading, Leading edge.

## I. INTRODUCTION

Studies conducted on composite materials and structures applicable in aircraft and aerospace structures are numerous [1-5]. In aircraft structure, composite material has been applied in various parts including fuselage, wing and tail. The main reason for the applications of composite material here is its high specific strength and stiffness that provide weight saving to the aircraft. However, most of these parts require accurate structural analysis as they are subjected to variety of loads that may lead to failures such as yielding, buckling, fatigue, parametric instability and impact fracture [6-10]. As shown in Fig.1, one important part of a wing of an aircraft is leading edge, a section commonly hit by impact loading that may be a consequence of bird impact or hail impact. These impactors come as a projectile that can hit and penetrate the wing structure commonly made of laminated composites. As a consequence, delamination or even fracture may occur to the structure and thus reducing its load carrying capacity [11].

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O. Nurihan, MJIIT UTM, Kuala Lumpur, Malaysia.  
Y. Aminanda, Faculty of Engineering, UTB Brunei Darussalam.  
J.S.M Ali, Kuliyyah of Engineering, IIUM, Gombak, Kuala Lumpur  
Z.A. Rasid, MJIIT UTM, Kuala Lumpur, Malaysia.  
H. Yahaya, MJIIT UTM, Kuala Lumpur, Malaysia.

It is thus significant to investigate the influence of impact loading on curved panels resembling the leading edge at the design stage of the aircraft wings.

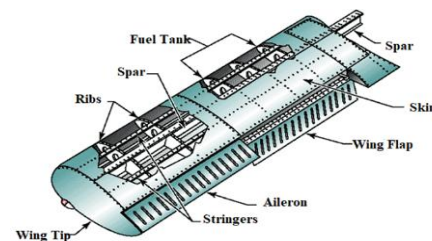


Fig. 1: An aircraft wing

The ability to resist impact load that may lead to fracture is characterised by the toughness of the structure. Toughness is a material property that give the amount of energy absorbed by the structure subjected to impact loading. For a leading edge of an aircraft to function well, it must be able to absorb a certain amount of kinetic energy i.e. it must have a minimum toughness. As such it is necessary to conduct impact test on all forms of leading edge to determine its toughness and parameters that affect the toughness.

Lo'pez-Puente et al [12] performed investigation on the breakage mechanisms of carbon/epoxy woven composite that occur during the penetration process of oblique ballistic impacts. The experimental works was successfully validated with numerical results. In a study, the curvature effect on the initiation of fracture of composite plates and shell was investigated [13]. Cylindrical shell structure was found to be affected more compared to flat plates. Furthermore, several FEM models have been developed to investigate the impact induced fracture of laminated composite structure [14-16]. Omar et al [17] conducted experimental study to investigate the effect of several parameters of curved panels such as curvature, thickness and angle of orientation on the impact behaviour of carbon fiber epoxy composite wing leading edge structure. The study discovered the significant effects that are actually provided by the parameters and the plots of the energy absorption per unit mass of structure against those parameters were given.

This study is to continue the work in [17] by applying finite element method in investigating the impact behaviour of carbon epoxy composite wing leading edge structure subjected to a solid low velocity projectile. Three shapes of leading edge panels were used: flat, semi-circular and semi ellipse while panels can be of

2, 4 and 8 layers to vary its thickness. The study on the effect of radius of curvature of the panels on the impact behaviour of the panels was also conducted.

II. METHODOLOGY

The material properties of the carbon-epoxy composite used here are given in this section. The dimensions of the curved panels that represents wing leading edge of aircraft are specified. The FEM procedures of modelling and analysis using LS-DYNA [18], a software owned by Livermore Software Technology Corporation (LSTC) are also elaborated.

A. Materials

The leading edge of the wing in this study is made of carbon-epoxy composite. Carbon fibre in woven fabric form was used here and, in the experiment, [17] performed. Tensile test was conducted on such the fabricated carbon-epoxy specimen to get the stress-strain plot such as shown in Fig. 2. In the LS-DYNA software applied in this study, the material model used is Mat-58 that is suitable for laminated composite with unidirectional layers and woven fibres. The Mat-58 applied the Hashin’s failure criteria while tolerating progressive failure analysis such that as the maximum effective strain is reached at certain elements, the specific elements are considered completely removed. For composites with woven fabrics and laminates such as in this study, the quadratic failure criteria are used such as:

Failure mode for tensile fibre ( $\sigma_{11} > 0$ ):

$$\left(\frac{\sigma_{11}}{X_T}\right)^2 + \left(\frac{\sigma_{12}}{X_S}\right)^2 = 1 \tag{1}$$

Failure mode for compressive fibre ( $\sigma_{11} < 0$ ):

$$\left(\frac{\sigma_{11}}{X_C}\right)^2 + \left(\frac{\sigma_{12}}{X_S}\right)^2 = 1 \tag{2}$$

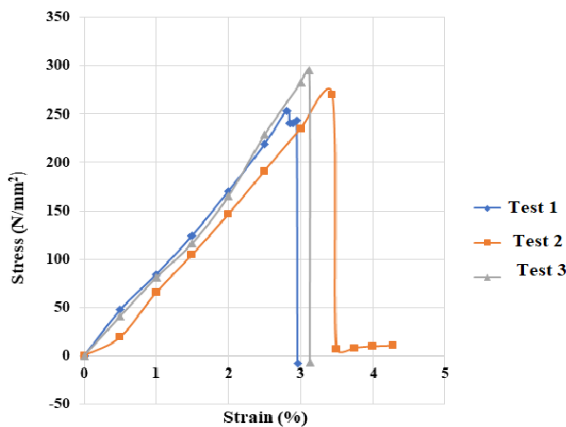


Fig. 2 : Stress-strain plot for the composite

On the other hand, the impactor is made of steel with slight adjustment in the steel density property to complement the experimental values [17]. Table I gives the material properties of the composite and the steel impactor.

Table 1 Material properties of composite and steel impactor

Material	Young’s modulus, E (GPa)	Poisson’s ratio, $\nu$	Density (Kg/mm <sup>3</sup> )
Composite	87.34	0.207	1.75(10 <sup>-6</sup> )
Steel	207	0.3	2.389(10 <sup>-3</sup> )

B. The dimensions

Following the experimental work performed by the authors [17], the leading edge applied here is of three forms: flat, semi-circular and semi-ellipse such as shown in Fig. 3. The effect of thickness of composite is studied by varying the composite number of layer such as the composites may have 2, 4 and 6 layers where thickness per layer is 1 mm. Depending on the number of layers, the selected angles of orientation of the composites are [0/90]<sub>n</sub>, [45/-45]<sub>n</sub> and [0/45]<sub>n</sub> where n will give the specified number of layers. In LS-DYNA, the leading edge panel was defined as SECTION\_SHELL as Mat-58 can only take shell element. By doing so, the thickness for each layer and the angle of orientation associated with each layer can be specified. Furthermore, the radius of the impactor ball is 10 mm. The impactor was defined as SECTION\_SOLID where the size of the steel impactor can be input.

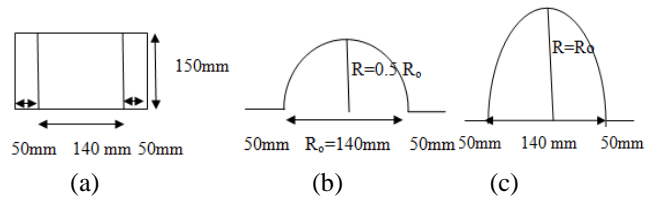


Fig. 3 The 3 forms of leading edge: (a) Flat plate (R = 0) (b) Semi-circular (R = 0.5R<sub>0</sub>), (c) Semi-ellipse (R = R<sub>0</sub>)

C. Boundary Conditions

The zero displacement occurring in all directions at the base of the leading edge panel was taken as the boundary condition of the structure. In LS-DYNA, the impactor is considered MAT\_RIGID as it is set to be un-deformed.

D. Energy Absorption

Through finite element analysis (FEA), the graph of displacement against time, velocity against time and force against time can be determined. The following formula is used to calculate the energy absorbed, E :

$$E = \frac{1}{2} * m * (v_i^2 - v_f^2) \tag{3}$$

where m is the mass of the composite, v<sub>i</sub> and v<sub>f</sub> are the initial and final velocities of the ball.

III. RESULTS AND DISCUSSION

Here, the results from the FEA is first validated with results from the experiment performed earlier. Following that, the results on the impact behaviours of the three forms of the trailing edges with varying radius of curvature and thickness are elaborated.



**A. Validation**

The validations of the FEA works were conducted by comparing to the author's previous experimental study [17]. The validation is performed on composites with the following specifications: 1. Flat plate,  $R = 0$  having 2-layer with angle of orientation,  $[0^\circ/90^\circ]$  2. Semi-circular plate,  $R = 0.5R_0$  having 4-layer with angle of orientation,  $[0^\circ/90^\circ]_2$ . While Fig. 4(a) shows a different numerical and experimental plots corresponds to flat leading edge due to inevitable problems of noise and vibration of machine at the time of experiment, for leading edge with semi-circular form in Fig. 4 (b), there is a good correlation in the graph of energy absorbed vs time corresponds to numerical and experimental work. The graph shows that increasing the impact time, the absorption of energy increases as well, in a non-linear fashion.

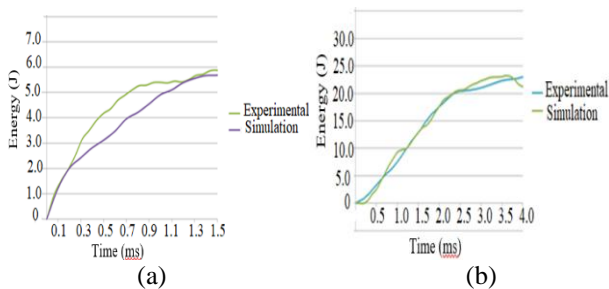


Fig. 4 : Experimental validations for leading edges with (a)  $R = 0$ , 2 layers,  $[0/90]$  and (b)  $R = 0.5R_0$ , 4 layers,  $[0/90]_2$

**B. Impact behaviour of composite leading edge panels**

The impact behaviour of composite subjected to low velocity impact is characterised by the deformation of the composite material until the point of fracturing. A material deforms as it absorbs increasing amount of energy while stress is created in the material. As the deformation is increased, stress is increased as well up to a certain level of energy absorbed, fracture starts to occur when the material cannot take the stress anymore. This is the point when the striker passes the specimen. Fig.5 shows the striker that passes the semi-circular, 2 layer specimen with  $[0/90]$  orientation angle. The amount of energy absorbed before fracturing, can be calculated using Equation (3) based on the graphs in Fig. 6. In Fig.6, the highest energy occurs at the beginning of the horizontal energy and velocity line that shows the specimen has been cut. In this case, the maximum energy absorbed is 28.1042 J.



Fig.5: Deformation of the semi-circular, 2 Layers,  $[0/90]$  specimen

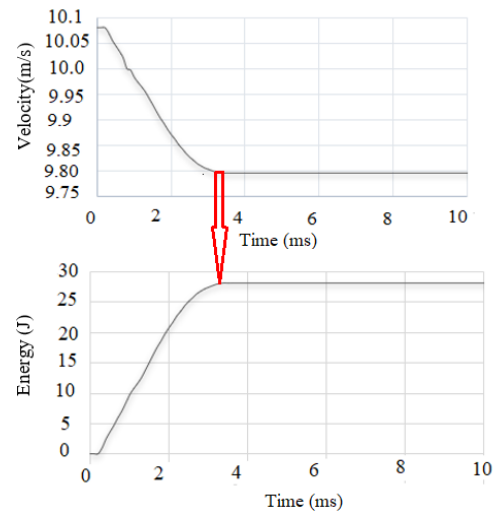


Fig. 6: The velocity and energy absorbed in the deformation of the semi-circular, 2 layers specimen

Increasing the thickness to 8 layers of the semi-circular specimen with  $[0/90]_4$  stacking sequence, the deformation process can be seen in Fig. 7 and Fig.8. From Fig. 8, it can be seen that it takes longer time of 3.9 s for the striker to fully penetrate the specimen. The velocity becomes constant at 9.797 m/s where the energy absorbed can be calculated as 31.28889 J.

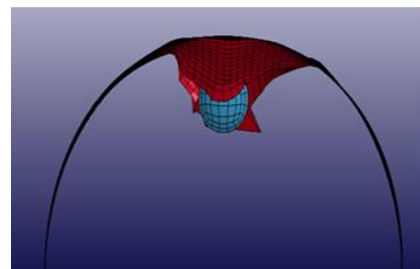


Fig. 7: Deformation of semicircular, 8 layers,  $[0/90]_4$  specimen

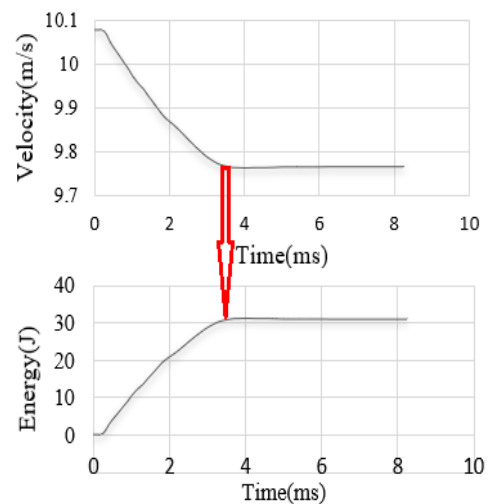


Fig. 8 The velocity and energy absorbed in the deformation of the semi-circular, 4 layers specimen

C. The radius of curvature effect

This study is to investigate the influence of radius of curvature of the leading edge panels on the impact behaviour of the panel. The radius of the leading edge was varied from 0 to 140 mm. The thickness and stacking sequence are fixed at 4 layers and [0/90]<sub>2</sub> respectively. Table II shows the energy absorbed per unit mass by specimens having varying radius of curvature while Fig. 9 gives the effect of radius of curvature of the leading edge on the specific energy absorbed.

Table II: Influence of radius of curvature of leading edge specimen on its energy absorption

Radius (mm)	Energy (J)	Specific energy (J/Kg)
0	2.67	20.6
35	10.95	71.2
70	23.09	132.7
105	20.7	100.48
140	17.1	71.8

From Figure 9, conclusion can be made that the increase of radius of curvature will also increase the specific energy. The maximum point occurs at radius, R = 70 mm and the specific energy starts to reduce following the maximum point. The highest energy per unit mass for 4 layers composite is 132.7 J/Kg.

D. The influence of panel thickness

This study is to investigate the effect of thickness of the leading edge panels on the impact behaviour of the panel. In this study, thickness of leading edge panels were varied while the stacking sequence of [0/90]<sub>n</sub> with R = 70 mm was used. In tabular form, the effect of panel thickness on the energy and specific energy absorbed can be seen in Table III. It shows that even though the energy absorbed is increased as the thickness is increased, the specific energy is decreased. This makes sense because the mass increases doubly moving from 2 to 4 and to 8 layers while the energy increases at lower rate compared to the rate thickness is increased. Fig. 10 shows clearly that specific energy decreases with the increase of the thickness of the specimen.

Table III: The effect of thickness on the specific energy

Thickness	Energy (J)	Specific Energy(J/Kg)
2 layers	20.07	230.69
4 layers	23.09	132.7
8 layers	31.08	89.31

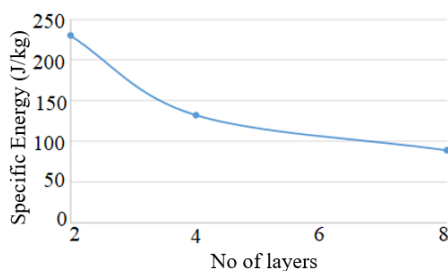


Fig. 10: Specific Energy Vs Thickness

Comparing to the experimental results [17], the energy absorption trend obtained through FEA is similar where the energy absorbed was increased with the increase of thickness.

E. Influence of Stacking Sequence

The effect of stacking sequence of the leading edge panels on the impact behaviour of the panels is given here. Since the radius of curvature for the highest energy absorption is R = 70 mm, the FEA computations for different stacking sequences are taken for model with R = 70 mm. Table IV shows the change of energy absorption capability of the leading edge panels with fibre orientation while Fig. 11 shows the plots of energy per unit mass against thickness of the leading edge panel. It can be seen from the plots that the highest energy absorbed is provided by the [0°/45°] configuration and followed by the [45°/-45°] and [0°/90°] stacking sequences.

Table IV: The change of Energy Absorption Capability with Angle of Orientation

Thickness	Fiber orientation	Energy (J)	Specific energy (J/Kg)
2 layers	[0°/90°]	20.07	230.7
	[45°/-45°]	22.6	259.8
	[0°/45°]	21.5	247.13
4 layers	[0°/90°]	23.2	133
	[45°/-45°]	27.4	157.5
	[0°/45°]	38.17	219.36
8 layers	[0°/90°]	31.3	89.9
	[45°/-45°]	40.4	116.1
	[0°/45°]	45.2	129.88

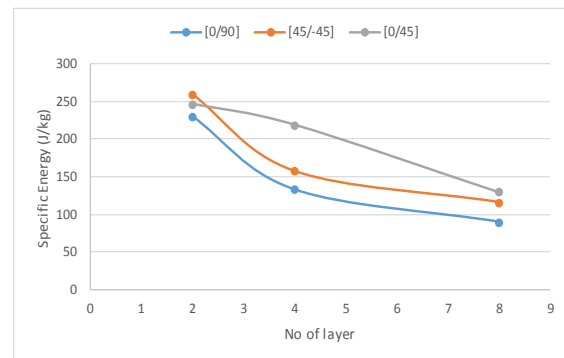


Fig.11: The effect of thickness on specific energy of the leading edge panels

IV. CONCLUSIONS

The impact behaviour of laminated composite panels exerted by steel ball impactor was studied numerically using the LS-DYNA FEM software. The composite panels can be of three forms i.e. the flat, the semi-circular and the semi-ellipse panels were representing the leading edge of aircraft. Material type Mat-58 in LS-DYNA was employed. Validations of the numerical work were successfully conducted based on experiment results for semi-circular panel having 4 layers with [0°/90°]<sub>2</sub>

orientation. Following that, several studies were performed to investigate the influences of panel thicknesses, angle of orientation and radius of curvature on the impact behaviour of the composite panels. It was found that panel with  $R = 70$  mm gives the highest energy absorption. Thus the radius,  $R = 70$  mm is the optimum radius of curvature for the range of specimen tested in this investigation. For the same curvature with different thickness, the plots of the energy per unit mass against thickness shows the reduction in specific energy absorption as the thickness is increased. Lastly, for the effect of angle of orientation, it can be said that the optimum stacking sequence is found to be for the combination of  $[0^\circ/45^\circ]$  stacking sequences

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### AUTHORS PROFILE



**O. Nurihan** was born in Pulau Pinang, Malaysia in 1986. She received the BSc in Mechanical Engineering (Aerospace) in 2010 and MSc in Mechanical in 2016 from IIUM, Gombak, Kuala Lumpur. She started her work in aviation industry as Quality Assurance Executive

in 2011. As a Quality Assurance personnel, she is certified Lead Auditor by International Register of certified Auditors (IRCA). Currently, she is pursuing her doctoral study at MJIIT, UTM Kuala Lumpur under Department of Mechanical Precision Engineering. Her current research interest includes the modelling of Carbon nanotubes. Previously, her research works was about the impact load on the composite structure.



**Y. Aminanda** was born in Bandung, Indonesia in year 1967. He received his BSc in Mechanical Science in Universite Paul Sabatier Toulouse, France. His doctorate study was in ISAE - ENSAE / SupAero Toulouse, France in Mechanical Engineering/Aerospace field. Previously, he was a Professor in Kuliyah of

Engineering, IIUM,

Gombak. Currently he is a Professor and also appointed as QA coordinator for Mechanical Engineering Program Area, Faculty of Engineering, at Universiti Teknologi Brunei, Brunei Darussalam. He is also a member of the Chartered Engineer (C.Eng UK), Engineering Council, United Kingdom, Member of the Institution of Mechanical Engineers (MIMEchE UK), United Kingdom, Member of AIAA ( American Institute of Aeronautics and Astronautics) USA, Member of AMAC ( L' Association pour les Materiaux Composites / Composite material association) France and Editorial board of journal of transport system engineering. His current research interests include Composite / sandwich Structure, Finite element Analysis/ CAE, Structural Analysis, Stress and Strength analysis, Impact / Crash on structure, Aircraft Structure and Design, Aeronautic Engineering and Airplane Certification. More than 60 research articles were published by him.



**J.S.M Ali**, a graduate of the Indian Institute of Technology Madras (IITM), obtained both his M.Tech. and Ph.D in the field of Aerospace Engineering in 1993 and 1998 respectively. He started his career in 1996 as a Scientist-B in Defence Research and Development Organisation, India. He joined B S Abdur Rahman

University, Chennai, India in 1998 as a senior lecturer in the Department of Mechanical Engineering and from 2003 to date, he is serving as an Assistant Professor at the Department of Mechanical Engineering, International Islamic University Malaysia. His area of specialization includes Natural Science, Material Sciences, Advanced Materials Engineering Tech (Applied Sciences And Technologies), Aerospace ( Aerospace Structures), Engineering Tech (Applied Sciences And Technologies) Engineering Materials, Information, Computer And Communications Technology (ICT) Software and Computer Aided Software Engineering (CASE) Tools. His current research project is about new Finite Element for Analysis of Smart Laminated Circular Cylindrical Shells under Hygro-Thermo-Electric-Mechanical Loads. To date, more than 50 articles including conference papers and more than ten book chapters was published by him.



**Zainudin A. Rasid** earned his PhD in Aerospace Engineering from the Universiti Putra Malaysia, Serdang in 2013. He is currently working as a senior lecturer at the UTM, Kuala Lumpur. He has published

more than 50 papers in forms of journal and conference proceeding mainly in the area of buckling and vibration of composites and smart composites and CNT composites and rotordynamic of shaft. His current areas of research are the composite material, carbon-nanotube reinforced composite, green composite, smart material, stress analysis, buckling and post-buckling analysis, plasticity,

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numerical method and finite element method. Dr. A. Rasid is a member of the Malaysian Board of Engineers.



**Hafizal Yahaya** finished his first degree in Physics from the Universiti Kebangsaan, Malaysia. He obtained his Master of Science in Information Technology in 2005 and his PhD in Material Science from the Kyushu University, Japan. He is currently the Senior Lecturer at the Universiti Teknologi Malaysia, Kuala Lumpur,

Malaysia

Dr. Yahaya has published more than 50 journals and conference proceeding papers. His research interests cover areas of thin film growth, epitaxial growth, semi-conductor nano-structure and bio-sensor.